

# Model Based FDIR process with Capella and COMPASS

Results of a CNES / TAS-F study and way forward

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MBSE ESA WORKSHOP – DEC 2016



- Context
- Study objectives
- TAS-F background
- Model Based Process
- Prototyping and use-case
- Way forward

## Context

## Previous R&T CNES study conclusions

- Evaluation of FDIR formal verification
  - Thanks to Model Simulation
  - Thanks to Model Checking techniques
- State of the art and experimentations conclusions :
  - Promising techniques to assist Spacecraft Safety, Dependability and FDIR engineers
  - Increasing efficiency and performance of existing tools
- However, rarely applied on programs
  - Not only in Space domain, but also in other domains.

## Why ?

- Need for detailed knowledge on the methodology and hands-on practice by prospective users.
- Big effort required to build a formal model of the system
- Insufficient means or methodology to ensure that the modelled system matches the real system
- Scarce understanding of properties to be proved
- Tools with shortcomings in ergonomic and interoperability with other engineering environments.

## How to ?

- Provide domain analysis (RAMS, FDIR, ...) based on formal methods / languages integrated to tools the user knows (engineering environments).
- Share modelling effort between engineering and Safety/FDIR teams
- Ensure that the modelled system matches the real system
- Clearly specify which properties need to be proved

## Study objectives

## CNES has initiated a study with the following objectives

- Define a Safety/FDIR process for Space Domain starting from the state of the art of other domains (aeronautical domain, transportation, ....)
- Identify the most suitable tools
- Experiment on a small case-study
- Elaborate some recommendations

## Timeline : 2013 - 2014

Purpose :

- Increase efficiency of safety analysis
- reduce the FDIR validation & verification costs



## TAS-F background

## Beginning of Model Driven Engineering:

- Slow & high-effort deployment of modeling techniques
- COTS are not well adapted to industrial needs
- Tool vendor dependencies are too constraining



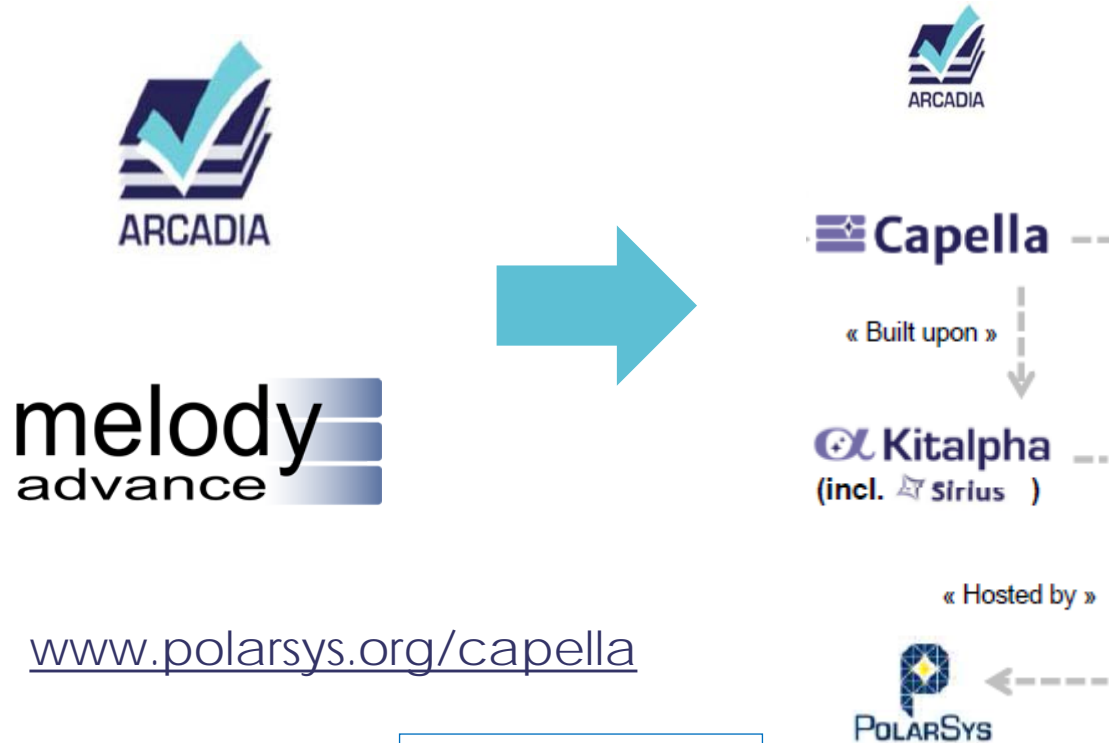
## THALES strategy:

- Define a method: ARCADIA
- Develop dedicated tooling: Melody Advance
  - Specified, designed & developed from operational needs
  - With the necessary capabilities (allow for quality and productivity, user-friendly, permits early validation, performance & scalability, suitable for configuration management and collaborative engineering, ...).
  - Applicable to every domain (Aeronautical, transportation, communication, ...)

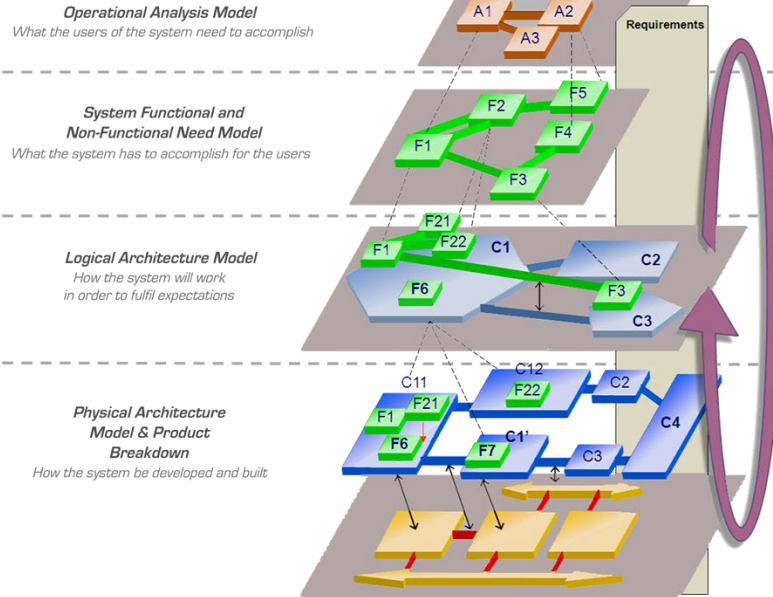


# TAS-F background – Melody Advance / Capella (1/)

## OSSing Melody Advance: public name is Capella



## Melody Advance / Capella



Satellite model

Avionics models

OBSW / Eqts models

Different scopes and purposes for modeling activities.

## ESA studies :

### > COMPASS

- develop a toolset for evaluation of system-level correctness, safety, dependability, and performance (performability) of the on-board computer-based systems.

### > COMPASS GRAPH

- Develop a graphical editor for SLIM models.

### > AUTOGEF

- Development of the Automated Model Generation Toolset for FDIR (AUTOGEF) as an add-on to the COMPASS Toolset, and definition of the associated methodology. (Synthesize FDIR diagnosis and controllers in SLIM model for an given system).

### > FAME (Failure and Anomaly Management Engineering )

- Definition of the FDIR development methodology and associated V&V process, and development of the Failure and Anomaly Management Engineering (FAME) Environment as an extension to COMPASS toolset.

### > FDI AOCS

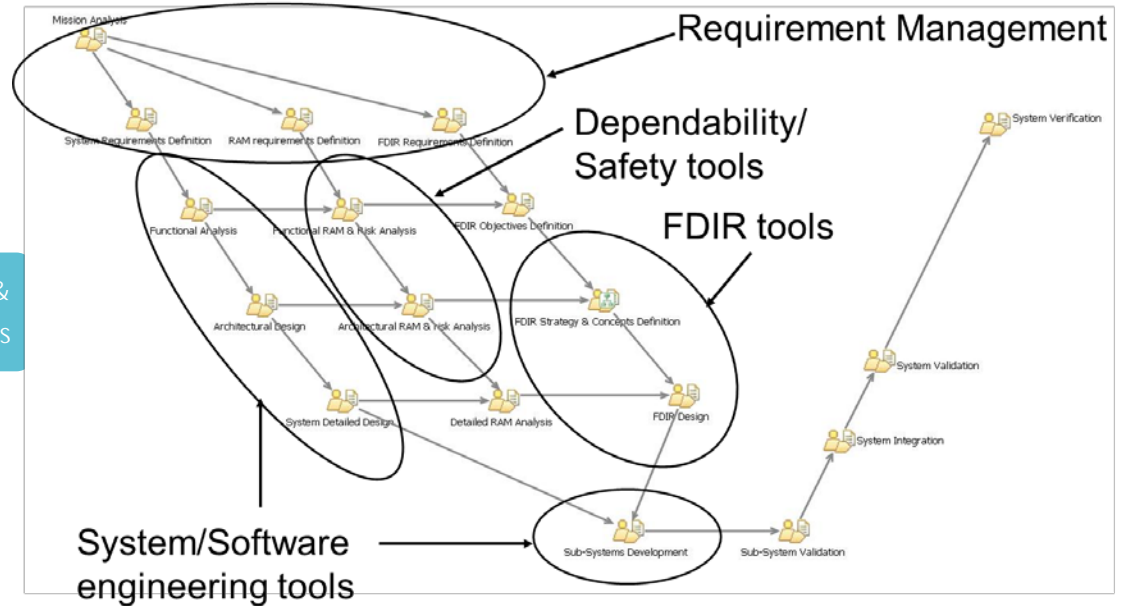
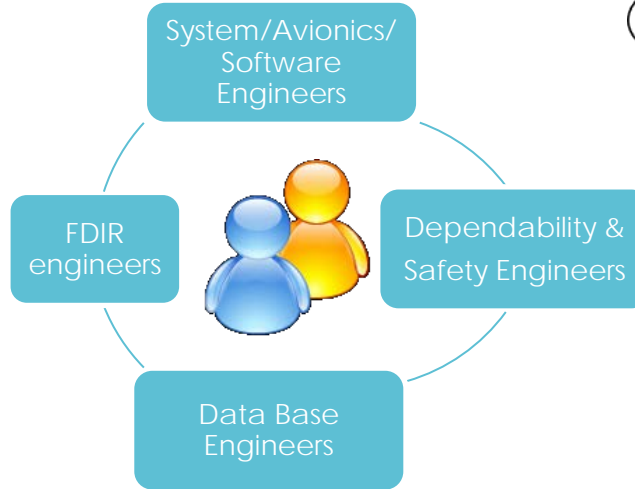
- Improvement of AOCS, FDIR & Avionics for compliance with LEO de-orbitation new requirements

## Model Based Process

# Model Based Process

## Proposed approach

- Identification of roles and activities => the process
- Identification of tools



# Model Based Process

## Proposed activities :

- Requirements analysis
- Functional /logical Analysis
  - System Functional Analysis
  - System Dependability / Safety analysis
  - Consolidation of the functional analysis
  - RAMS analysis at functional level
- System Physical Design
  - System design at physical level
  - System Dependability / Safety analysis at physical level
  - Consolidation of the System Physical Design
- FDIR development
  - FDIR requirement analysis
  - FDIR objectives definition
  - FDIR concepts and FDIR strategy specification
  - FDIR design
  - FDIR implementation
  - Final FDIR V&V

Aligned with ESA  
study results

## Proposed tools :

- Requirements Management tool (e.g. DOORS)



- Functional Analysis
- Logical / Physical Design
- Dependability / Safety analysis
- RAMS analysis

melody  
advance

Capella

COMPASS

- FDIR specification
- FDIR design
- FDIR Verification

FDIR Editor

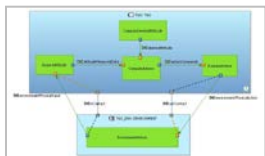
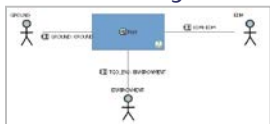


## Prototyping and use-case

# Prototyping and use-case

## Exomars TGO case study

### Functional Analysis



### Functional Hazard Analysis

- **Omission** : the function is not performed when demanded.
- **Commission** : function is performed when not demanded.
- **Late** : The function is performed later than required.
- **Early** : The function is performed earlier than required.
- **Value** : The output value is not correct

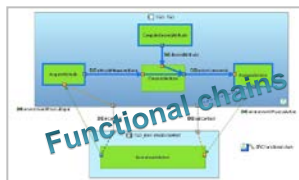
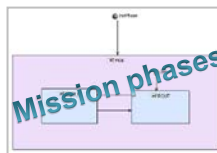
Guide words

Functional Hazard ID	Functional Hazard Name	Functional Hazard Description	Functional Hazard Severity
H21	Attitude Control Loss	"The spacecraft is unable to maintain attitude. The battery can become unable to avoid space debris (1)"	Loss of system: catastrophic (1)
H22	Erroneous Attitude during MOI phase	"The spacecraft attitude is not correct and makes the spacecraft miss the critical Mars orbit insertion manoeuvre"	Loss of system: catastrophic (1)
H23	"The spacecraft attitude is not correct what can alter the data transmission to ground or the power supplying chain"		mission degradation (2-3-4)

- H21 → REQ1 "The Probability to lose the spacecraft attitude during mission shall be lower than 10<sup>-6</sup>"
- H22 → REQ2 "In MOI phase, the Control Functions shall recover in less than X seconds"
- H23 → REQ3 "Control Function shall be robust to any single Failure"



### Safety objectives allocation on FA

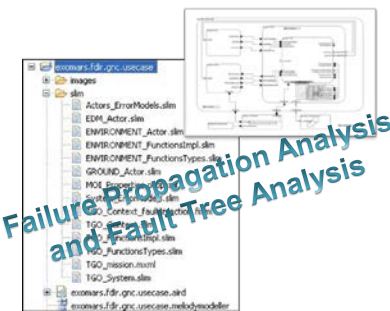


Worst case scenarios

### RAMS analysis

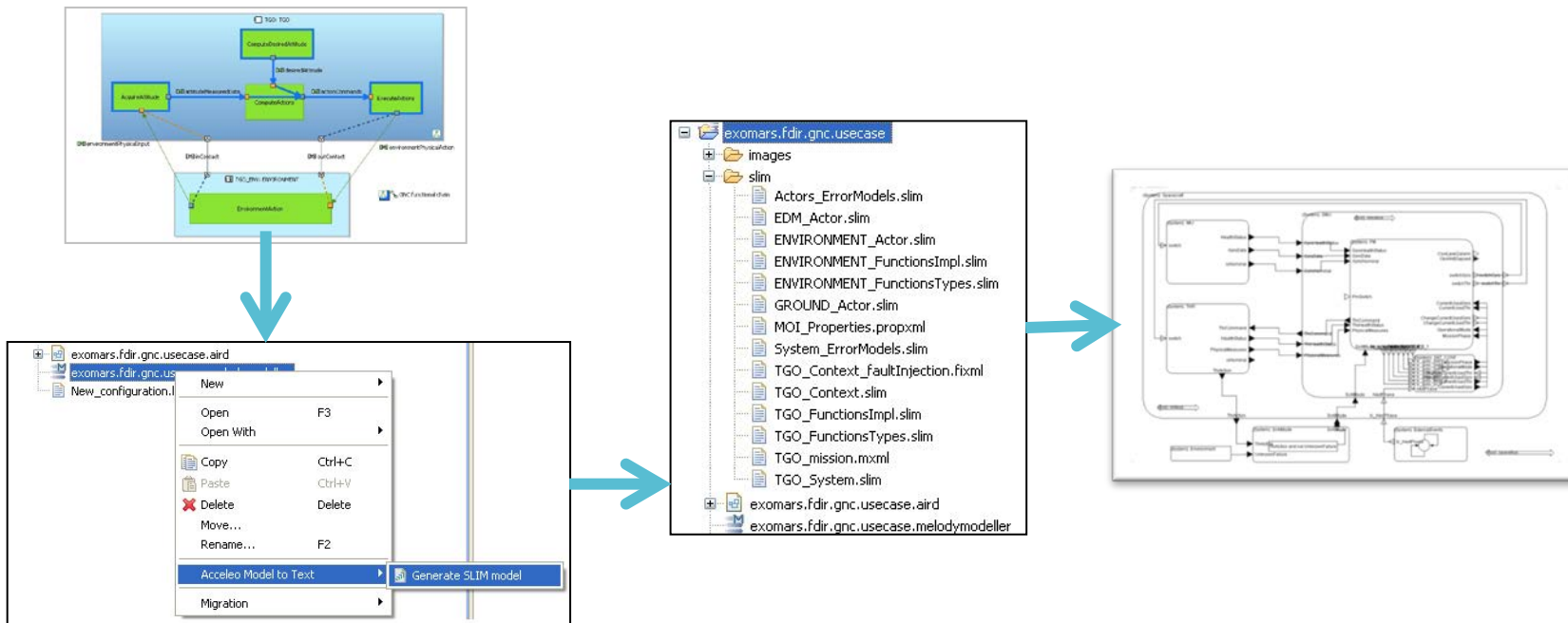
Mission Phase	Feared Event	Sub-Feared Event	Involved Unit
Mars Orbit Insertion	Critical Attitude Control	Compute desired spacecraft attitude function	Compute desired spacecraft attitude function
		Measure spacecraft attitude function	Measure spacecraft attitude function
		Compute Actions for desired attitude failure	Compute Actions for desired Attitude Function
		Execute Action Failure	Execute Action Function

Involved Unit	Failure Mode	Effects on output functional exchanges
Compute desired spacecraft attitude function	Bad desired attitude	BAD_DESIRED_ATTITUDE
Measure spacecraft attitude function	No measures	BIASED_MEASURES
	Biased measures	BIASED_MEASURES
Compute Actions for desired Attitude function	Error in action generation	ERRONEOUS_NO_ACTIONS
	Biased actions are produced	BIASED_ACTIONS
Execute Action Function	Erroneous actions are produced	ERRONEOUS
	No action is executed bad action is executed	NO_ACTIONS BAD_ACTIONS



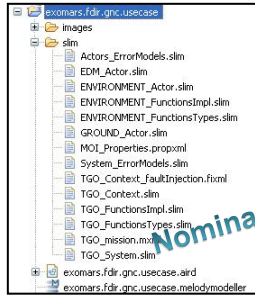
# Prototyping and use-case

## From Melody Advance/ Capella to SLIM models

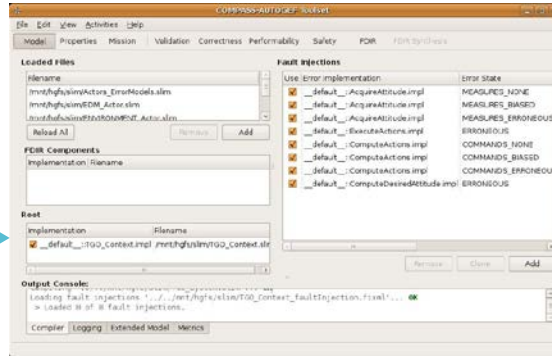


# Prototyping and use-case

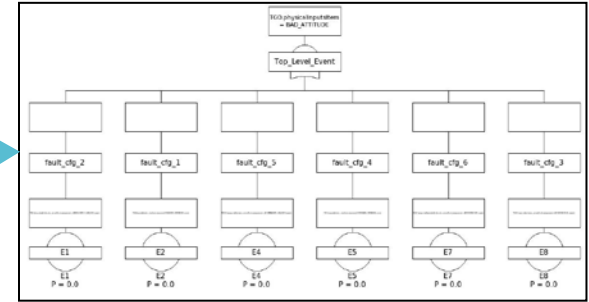
## COMPASS analysis at functional level



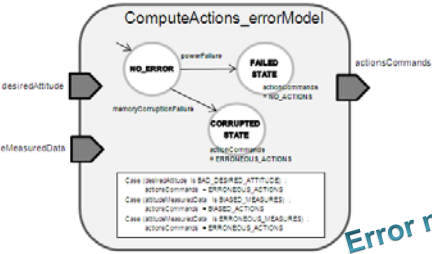
Nominal SLIM model



Fault injection within COMPASS



Fault Tree Analysis within COMPASS



Error models

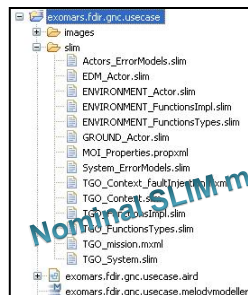
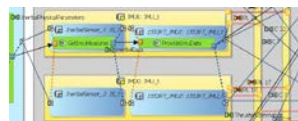
# Prototyping and use-case

## System design

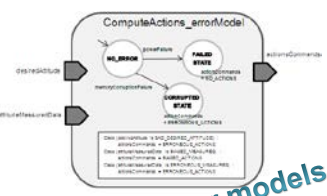
- Physical design
- FMEA based on Equipment Datasheet (EDS?)
- Modeling of Failure modes and fault propagation at physical level
- Fault Injection and COMPASS analysis : FMEA / FTA



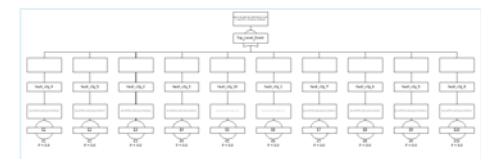
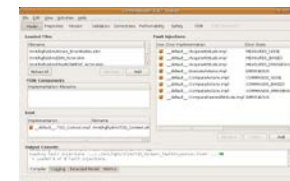
Product	Project/Phase	System/Subsystem/Equipment	Failure Mode	Failure Cause	Failure Effect	Severity	Control	Prevention	Correction	Remarks



Normal SLIM models



Error models



➤ Criticality analysis

ID	Item	Function	Failure Mode	Failure Cause	Impact	Failure Effects	Severity	Failure Analysis	SA	PE	CA	Comments/Remarks
F01	GPU	Process initial parameters	No resources	Internal failure	MCB	No resources, no sent to GPU	loss of attitude	absence of resources	3	2	4	switch to redundant
F02	GPU	Process initial parameters	Excess resources	an external sensor	MCB	resources are loaded	loss of attitude	Excess resources, blocked by a filter	1	3	2	reset equipment
F03	GPU	Process initial parameters	Erroneous Resources	Internal failure	MCB	resources are erroneous	erroneous attitude	No detection and attitude is affected by erroneous	3	3	4	switch to redundant

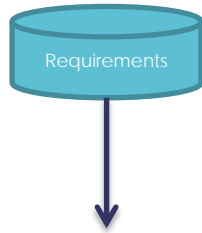
## Remarks

- The process should be iterative
  - Add new safety related functions (filter, detection, ...)
  - Specify new observables / commandable data
  - Add FDIR related components / mitigation / redundancy
  - Add cross-strapping
  - ...

# Prototyping and use-case

## FDIR

- FDIR objectives definition
- FDIR strategy definition



- **FDIR\_OBJ1:** Surviving shall be ensured for any single failure
- **FDIR\_OBJ2:** Achieve S/C manoeuvres for critical phase even in case of failure
- **FDIR\_OBJ3:** Fuel consumption shall be optimized and reconfiguration and equipment loss shall be minimized

- Classify failures
  - Detection level
  - Isolation level
  - Reconfiguration level



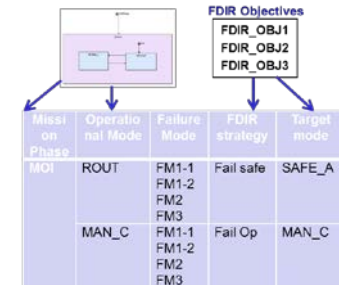
Fault	Detection level	Reconfiguration level
FM1-1	Level 0	Level A
FM1-2	Level 1	Level B
FM2	Level 2	Level B
FM3	Level 2	Level B

Detection	IMU	CSW
HW only	FM1-1	
Processor unit with SW		FM1-2 FM2 FM3

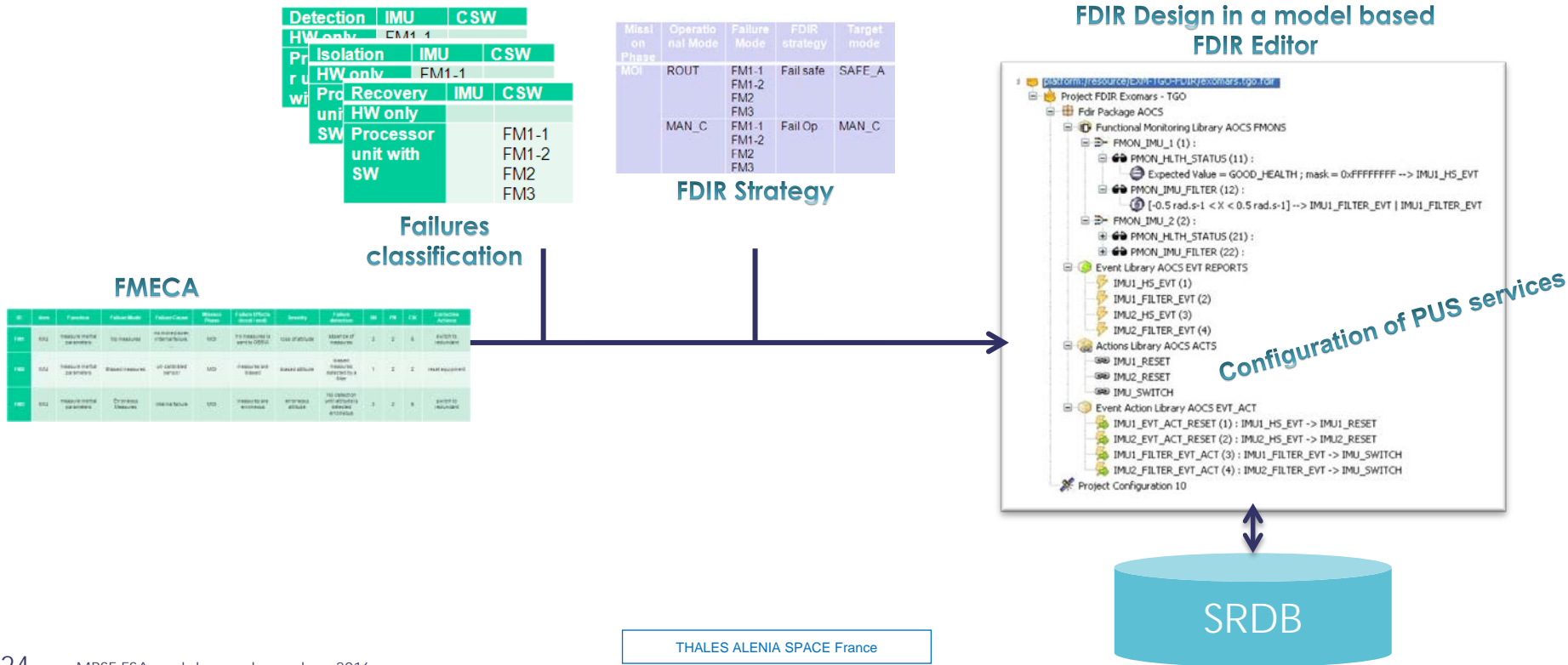
Isolation	IMU	CSW
HW only	FM1-1	
Processor unit with SW		FM1-2 FM2 FM3

Recovery	IMU	CSW
HW only		
Processor unit with SW		FM1-1 FM1-2 FM2 FM3

- Define reconfiguration strategies
  - Fail'Op
  - Fail'Safe
  - ...



## Configuration of PUS monitoring and action services based on models





## Way forward

# Way forward

## Step 1 : Deployment of Model Based practices in the Engineering process

- At architecture level
  - Requires a mature and already proven tool and methodology (like Capella and Arcadia)
  - Guidelines and validation rules should ensure semantics of the models
  - Models should be used to produce artefacts : specification / code / database / tests / ...
- At behaviour level:
  - Use of different formalisms to cope with different contexts (Matlab/Simulink, SDL, SLIM, TFGP, scenarios, ...)
  - Models should be used to produce artefacts : specification / code / tests / ...
  - Behavioural models should be coupled with architectural design models

## Step 2 : Define a Safety / Dependability / FDIR reference architecture

- Extend the Avionics Reference Architecture (ASRA) and the On-Board Software Reference Architecture (OSRA) with dedicated concepts and methodology
- Provide dedicated « viewpoints » in the engineering tools (Matlab/Simulink, Capella, SCM, ...)
- Investigate use of Electronic Data Sheets to support Failure Mode definition (at equipment level) and coupling with engineering models.
- Focus on production of artefacts (specification, code, configuration, ...)

## Step 3 : coupling Model Checking and Simulation tools

- Consolidate objectives : early validation of the system design (redundancy, cross-strapping, strategy, ...), generation of FMEA, FTA, Failure Propagation Analysis, ....
- Connect Model Checking & Simulation tools to Engineering tools : set-up model to model (M2M) transformations like Capella -> AADL/SLIM
- Map the system behavioural models to formal languages used for model checking and simulation (cope with synchronisation, timing aspects, ...)

# Way forward

## Classical approach :

